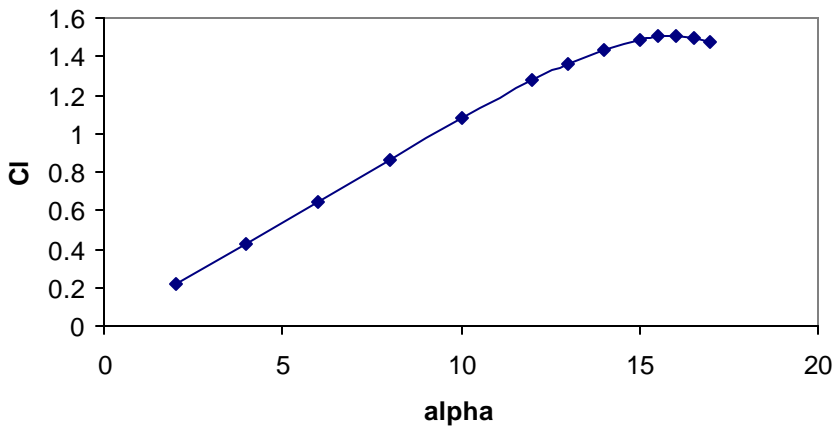
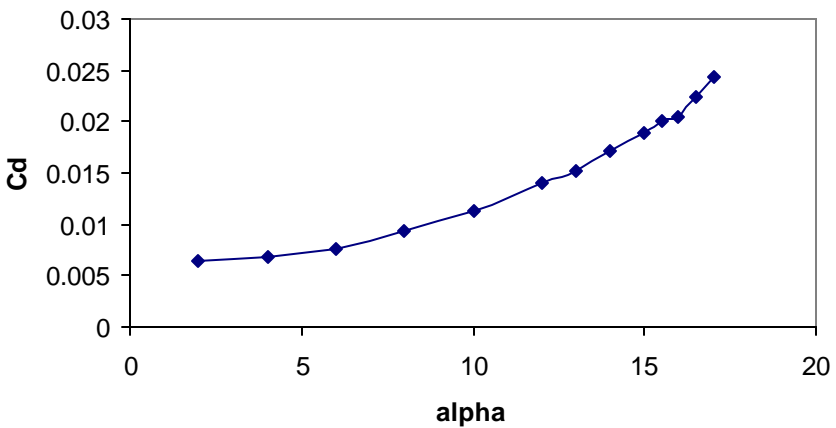


j	alpha	cl	cd	cm
1	2	0.21462	0.00645	-0.05132
2	4	0.42853	0.00678	-0.10219
3	6	0.64688	0.00757	-0.15469
4	8	0.86734	0.00931	-0.20815
5	10	1.07809	0.01136	-0.25702
6	12	1.27379	0.01398	-0.29942
7	13	1.36075	0.01519	-0.31614
8	14	1.43456	0.01714	-0.32817
9	15	1.48369	0.01885	-0.33068
10	15.5	1.50225	0.02	-0.33024
11	16	1.50439	0.02049	-0.32461
12	16.5	1.49441	0.0225	-0.31612
13	17	1.47769	0.02433	-0.30817

Lift Coefficient, $RL=4E6$, Mach=0.1



Drag Coefficient, $RL=4E6$, Mach=0.1



Momentum Coefficient, $RL=4E6$, Mach=0.1

