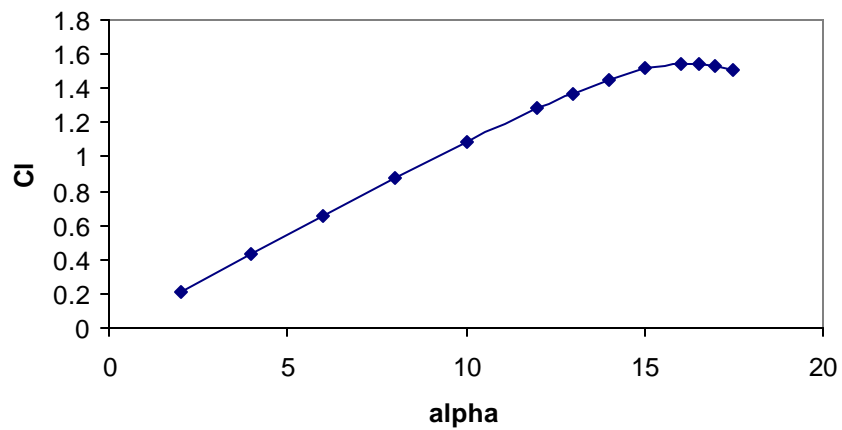
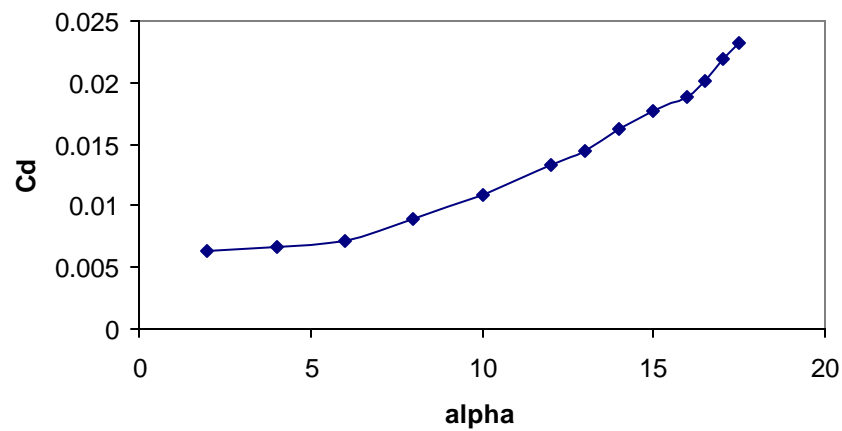


j	alpha	cl	cd	cm
1	2	0.21477	0.00636	-0.05136
2	4	0.42856	0.00662	-0.10215
3	6	0.64955	0.00719	-0.15585
4	8	0.87079	0.0089	-0.2095
5	10	1.08446	0.01092	-0.25953
6	12	1.28279	0.01332	-0.30278
7	13	1.37241	0.01452	-0.32033
8	14	1.45074	0.01628	-0.33378
9	15	1.51397	0.01762	-0.34138
10	16	1.54724	0.0188	-0.33864
11	16.5	1.54708	0.02005	-0.3322
12	17	1.53331	0.02199	-0.32274
13	17.5	1.50337	0.02322	-0.31044

Lift Coefficient, RL=5E6, Mach=0.1



Drag Coefficient, RL=5E6, Mach=0.1



Momentum Coefficient, $RL=5E6$, Mach=0.1

